

A HYBRID RECURRENT NEURAL NETWORK AND LONG SHORT-TERM  
MEMORY FOR SIMPLIFIED GENERAL PERTURBATIONS-4 MODEL IN  
ORBIT PROPAGATION

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## **DEDICATION**

This thesis is dedicated to my late parent, my beloved husband Mohd Nazrul Hanif, and my beautiful children Aqeela, Ameena, and Haqeel.

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In the name of Allah, Most Gracious, Most Merciful,

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## ABSTRACT

Orbit propagation is one of the critical science tasks used to determine and forecast the position and velocity of orbiting space objects such as satellites, mission-related debris, rocket bodies, and others. Developing an accurate orbit propagation model is vital to ensure uninterrupted operational planning and prevent any disrupted collisions or disasters. However, using the current orbit propagation model has limitations, and these reduce the ability for long-term forecasting. It has errors depending on various aspects like measurement error, space environment information that constantly changes, inherent uncertainty in the data used, and errors in the data processing. Although classical time series methods such as Holt-Winters can improve the orbit propagator's accuracy and efficiency, it requires changes in the components' probability distribution, causing complexity and computational burden for end-user. However, this method can achieve maximum performance through integration with other approaches. Deep learning techniques, the new field of research within machine learning, are recently explored to analyse and improve the Simplified General Perturbations-4 (SGP4) Model, the orbit propagation model commonly used by space operators. The improved model should minimize errors and maintain accuracy even if the propagation span increases. Therefore, this study examined the Recurrent Neural Network (RNN) and Long Short-Term Memory (LSTM) technique, a deep learning approach dealing with long-term time-series data. It can learn tasks and deal with complicated problems. Additionally, these learning techniques are a time series forecasting method that can improve models by capturing periodic data patterns by memorizing and learning from historical data. Thus, a hybrid RNN-LSTM SGP4 Model was developed. The performance and effectiveness of the improved model were evaluated and validated. As a result, this hybrid RNN-LSTM SGP4 Model improved more than 27% better than the SGP4 Model alone. It was also capable of being a reliable long-term time series forecasting model for space object data.

## ABSTRAK

Penyebaran orbit adalah merupakan salah satu tugas sains kritikal yang digunakan untuk menentukan dan meramalkan kedudukan dan halaju objek angkasa yang mengorbit seperti satelit, serpihan berkaitan misi, badan roket dan lain-lain. Membangunkan model ramalan orbit yang tepat sangat penting untuk memastikan perancangan operasi tidak terganggu dan mencegah sebarang pelanggaran atau bencana berlaku. Walau bagaimanapun, penggunaan model penyebaran orbit semasa mempunyai kekangan dan mengurangkan keupayaan untuk ramalan jangka panjang. Kekangan ini adalah disebabkan oleh pelbagai aspek, termasuk kesalahan pengukuran, maklumat persekitaran angkasa yang sentiasa berubah, ketidakpastian pada data yang digunakan, dan kesalahan dalam pemprosesan data. Walaupun kaedah ramalan siri masa klasik seperti Holt-Winters dapat meningkatkan ketepatan dan kecekapan penyebaran orbit, kaedah ini memerlukan perubahan pada taburan kebarangkalian komponen yang menyebabkan kerumitan dan beban pengiraan kepada pengguna akhir. Selain itu, kaedah ini boleh mencapai prestasi maksimum melalui integrasi dengan pendekatan lain. Terkini, teknik pembelajaran mendalam, iaitu satu bidang penyelidikan baharu dalam pembelajaran mesin telah diterokai untuk menganalisa dan menambah baik Model *Simplified General Perturbations-4* (SGP4). Model ini adalah model penyebar orbit yang biasa digunakan oleh operator angkasa. Model yang ditambah baik ini perlu meminimumkan ralat di samping dapat mengekalkan ketepatan walaupun di dalam jangka masa penyebaran yang meningkat. Oleh itu, penyelidikan ini mengkaji teknik Rangkaian Neural Berulang (RNN) dan Memori Jangka Pendek Panjang (LSTM), iaitu pendekatan pembelajaran mendalam yang boleh memproses data siri masa jangka panjang. Ia dapat mempelajari tugas dan menangani masalah yang rumit. Selain itu, teknik pembelajaran ini adalah kaedah ramalan siri masa yang dapat menambah baik model dengan menangkap corak data berkala dengan menghafal dan belajar daripada data sejarah. Oleh itu, Model RNN-LSTM SGP4 hibrid dibangunkan. Prestasi dan keberkesanan model yang dipertingkatkan ini dinilai dan disahkan. Hasilnya, prestasi Model RNN-LSTM SGP4 hibrid ini meningkat 27% lebih baik daripada Model SGP4. Ia juga mampu menjadi model ramalan siri masa jangka panjang yang boleh dipercayai untuk data objek angkasa.

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## LIST OF ABBREVIATIONS

ANN	-	Artificial Neural Network
ARIMA	-	Auto-Regressive Integrated Moving Average
BLC-IRK	-	Bandlimited Collocation Implicit Runge-Kutta
CADET	-	Covariance Analysis Describing Function Technique
CNN	-	Convolutional Neural Network
CPU	-	Central Processing Unit
DBM	-	Deep Boltzmann Machine
DBN	-	Deep Belief Network
DSST	-	Draper Semi-Analytical Satellite Theory
ECEF	-	Earth-Centred, Earth-Fixed
ECI	-	Earth Centred Inertial
EKF	-	Extended Kalman filter
ETS	-	Error, Trend, Seasonal
GA	-	Genetic Algorithm
GEO	-	Geostationary Orbits
GPU	-	Graphics Processing Unit
HEO	-	High Ellipticity Orbits
JLNELN	-	Joint Linear-Nonlinear Extreme Learning Network
LEO	-	Low Earth Orbit
LR	-	Linear Regression
LSTM	-	Long Short-Term Memory
MAPE	-	Mean Absolute Percentage Error
MEO	-	Medium Earth Orbit
NCA	-	Neighbourhood Component Analysis
NLR	-	Nonlinear Regression
NORAD	-	North American Aerospace Defense Command
ODEM	-	Orbit Determination for Extended Manoeuvres
PCE	-	Polynomial Chaos Expansions
P <sub>DL</sub>	-	Performance of Deep Learning
RAAN	-	Right Ascension of the Ascending Node

RMSE	-	Root-Mean-Square Error
RNN	-	Recurrent Neural Network
RSO	-	Resident Space Object
RQ	-	Research Question
SDPSO-ELM	-	Switching Delayed Particle Swarm Optimization
SGP4	-	Simplified General Perturbations-4
SSA	-	Space Situational Awareness
SST	-	State Transient Tensors
STK	-	Satellite Toolkit
SVM	-	Support Vector Machine
TDNN	-	Time-Delay Neural Network
TEME	-	True Equator Mean Equinox
TLE	-	Two-Line Element
UKF	-	Unscented Kalman Filter
UTM	-	Universiti Teknologi Malaysia

## LIST OF SYMBOLS

$a_E$	-	The equatorial radius of the Earth
$B^*$	-	The SGP4 type drag coefficient
$D, d$	-	Diameter
$e_o$	-	The "mean" eccentricity in epoch
$F$	-	Force
$i_o$	-	The "mean" inclination at epoch
$J_2$	-	The second gravitational zonal harmonic of the Earth
$J_3$	-	The third gravitational zonal harmonic of the Earth
$J_4$	-	The fourth gravitational zonal harmonic of the Earth
$I$	-	Moment of Inertia
$M_o$	-	The "mean" mean anomaly at epoch
$p$	-	Pressure
$r$	-	Radius
$ro$	-	Position
$Re$	-	Reynold Number
$vo$	-	Velocity
$XKMPER$	-	kilometres/Earth radii
$\Delta$	-	Minimal error
$\omega_o$	-	The "mean" argument of perigee at epoch
$\Omega_o$	-	The "mean" longitude of ascending node at epoch
$\dot{n}_o$	-	The time rate of change of "mean" motion at epoch
$\ddot{n}_o$	-	The second time rate of change of "mean" motion at epoch

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# CHAPTER 1

## INTRODUCTION

### 1.1 Overview and Motivation

Space situational awareness (SSA) is a critical issue affecting the space industry and national security. SSA refers to the ability to observe, characterize and forecast the properties of natural and artificial objects that orbit the Earth. It also aims to prevent collisions, identify untracked objects, and ensure security for future missions (Oltrogge and Cooper, 2020; Pelton, 2019; Choi *et al.*, 2017). One of the critical technical discussions in the SSA is that space objects propagate state uncertainties due to dynamic and nonlinear environmental factors (Park, 2016). Besides that, the lack of information such as the space environment information and space object characteristics causes the current orbit propagation model unable to achieve the accuracy required for proper operational planning and avoid space object collisions (Tarran, 2021; Peng and Bai, 2018a).

Every year, the number of resident space objects (RSOs) orbiting the Earth increases and indirectly increases the conflict between RSOs (Lu *et al.*, 2020; Luo and Yang, 2017; Pelton and Jakhu, 2017; Park, 2016). It consists of satellites, mission-related debris, rocket bodies, etc. The growth is due to many countries scrambling to explore space for various purposes such as communications, remote sensing, scientific mission, security, defence, and many more. The number of space objects larger than 10cm is now approaching 21,000, an object of between 1 and 10cm is estimated to be around 500,000, and for objects smaller than 1cm, it is to be over 100 million (Peng and Bai, 2018a). Moreover, there is a large amount of debris in the orbits around the Earth, all of which endanger space assets and society (Tarran, 2021; Gambi *et al.*, 2018; San-Juan *et al.*, 2017; Lim, 2015). Figure 1.1 shows the number of objects in Earth Orbit by object type increasing yearly.

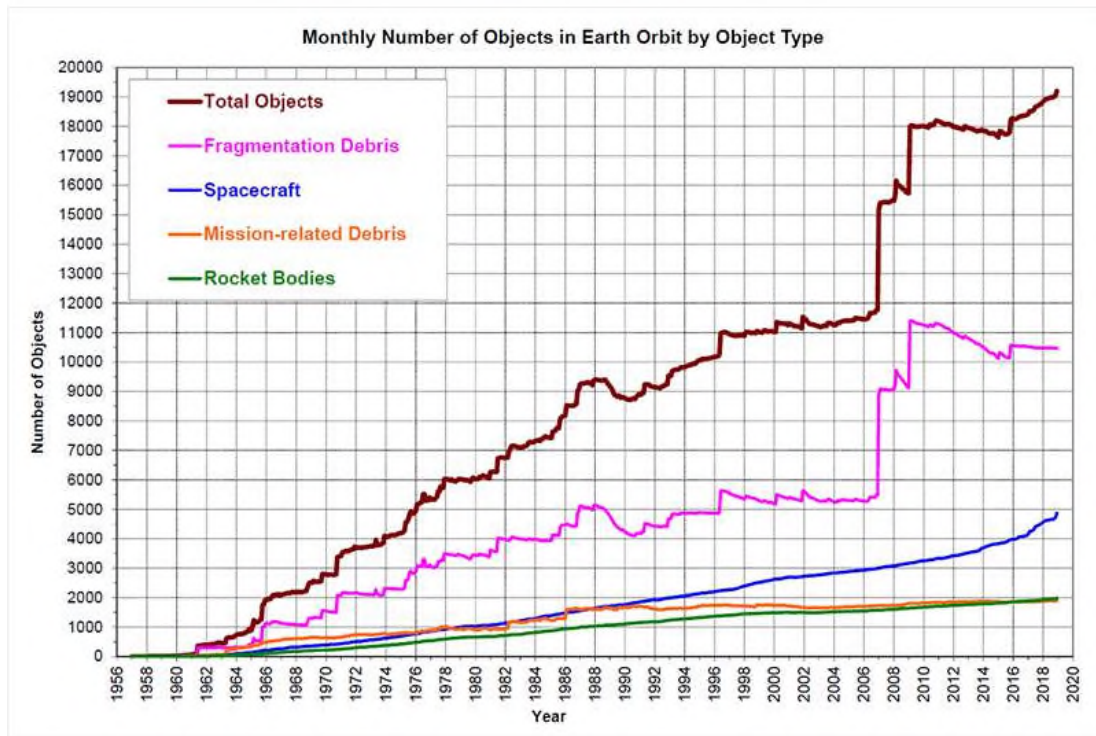


Figure 1.1 Yearly Number of Objects in Earth Orbit by Object Type (Source: <https://orbitaldebris.jsc.nasa.gov/modeling/legend.html>)

Some incidents involving space objects have also occurred, such as the February 2009 collision involving the U.S. Iridium communication satellite and Russian Cosmos 2251 communication satellite, as well as the threat of RED threshold late notice conjunction with the International Space Station (ISS) from the "25090 PAM-D" debris (Seong et al., 2017; Kelso, 2012; Jakhu, 2010; Bergin, 2009). One of the leading causes of these incidents is the orbit propagation model's ability to obtain accurate information about the satellite's position (Peng and Bai, 2019). The physics-based prediction failure also arises from a lack of information such as the space object's state, the initial time, the environment information, the intent information, etc. Figure 1.2 shows the Iridium 33 and Cosmos 2251 collision illustration view, which created more debris.



Figure 1.2 Illustration view of Iridium 33 and Cosmos 2251 Collision (Source: <https://celestrak.com/events/collision/>)

On April 2, 2018, the Chinese space station Tiangong-1 (Heavenly Palace) was declared out of control by Chinese authorities and re-entered the atmosphere (Goswami, 2018; Danner, 2018). Later, the incident involved a Long March 5B Rocket, also owned by China, had entered atmospheric space out of control (Zannoni, 2022). Therefore, the authorities should continuously track this space object to ensure its re-entry into the atmosphere can be immediately detected to prevent any accident from happening. Unfortunately, the related information is usually only obtained after the incident occurred. Thus, having the orbit propagation model that can forecast for long-term horizon will help affected parties have proper planning operations and take extra precautions to prevent collisions and observe the space objects orbiting the Earth. This orbit propagation model will also help identify possible crash locations to ensure no casualties. Countries with no space object information details can also take preventive measures using this orbit propagation model. Therefore, further analysis needs to ensure that no adverse events occur, especially collisions between the operating satellites resulting in the loss of millions of dollars.

Besides that, an accurate orbit propagation model is crucial to maintain catalogue objects' growth, conduct collision prevention assessments, and address satellite missions currently operating in orbit (Reiland *et al.*, 2021; Bradley, 2015; Bennett *et al.*, 2013). These also drive the need to improve the current orbit propagation model.

The need for high accuracy and efficiencies prompted various studies to renew the orbit propagation method (Petit *et al.*, 2021; San-Juan *et al.*, 2017). One of the methods proposed is to propagate uncertainty by capturing the absence of dynamic information. Hence, it remains a challenge to forecast space objects. Among the challenges is our understanding of the space environment is limited. The information about the space object is not accurately updated, manoeuvring of a spacecraft could be unavailable, surveillance resources are expensive, and measurements are sparse and noisy.

The Simplified General Perturbations-4 (SGP4) Model is the orbit propagator model commonly used by the satellite operator as its code and input data; the Two-Line Elements (TLE) are publicly available and accessible by everyone (Driedger and Ferguson, 2021; Peng and Bai, 2020; San-Juan *et al.*, 2017). However, the TLE data must be updated to have an accurate result, or else the SGP4 Model error can be 1 km and grow  $\sim 1-3$  km per day (Romano *et al.*, 2021; Riesing, 2015; Vallado *et al.*, 2006a). The North American Aerospace Defense Command (NORAD) provides the TLE data daily, but its accuracy is valid for 2- 3 days (Abay *et al.*, 2021; Romano *et al.*, 2021). Therefore, the SGP4 Model is unreliable for long-term forecasting horizons, and it shall be improved to increase its performance with minimal risk and cost.

## **1.2 Background of the Problem**

The growth of space objects increases the risk of collision among those objects and, at the same time, poses a danger to space assets and humans. Also, the satellite communication window, especially in the Low Earth Orbit (LEO), is concise at about 7 to 12 minutes. This limited time is valuable for scheduling satellite operations such as satellite health checks, uploading schedules, and downloading mission data. The planning of this satellite schedule is prepared based on the time determined by the orbit propagator. However, there are distortions and noise in the orbit propagation caused by orbit perturbations such as earth gravity, atmospheric drag, multi-body gravity, and solar radiation pressure. Therefore, the orbit propagator becomes inaccurate and needs



to be improved to minimize the error, reduce this risk, and ensure future mission safety.

Figure 1.3 illustrates the overview of the orbit propagation approaches used.

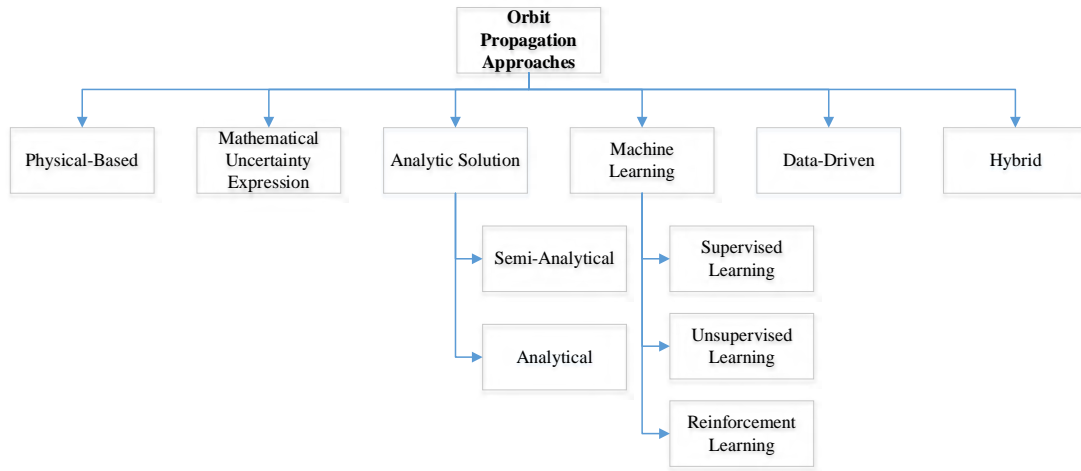


Figure 1.3 Overview of the Orbit Propagation Approaches

The satellite operation typically uses the physical-based approaches for orbit propagation. However, this approach's success requires knowing the space object at the beginning of trajectory calculations, environmental information, and manoeuvre the objects (Peng and Bai, 2018a). Meanwhile, the understanding of the space environment is limited. Furthermore, the information on space objects is not updated accurately; for example, a satellite owned by another country is not accessible if the owner is unwilling to share the information. Also, the current surveillance resources are limited and costly (Peng and Bai, 2018a).

There are various mathematical uncertainty expressions used for SSA, such as Gaussian, Polynomial Chaos Expansions (PCEs), State Transient Tensors (STTs), and Taylor Series Polynomial (Park, 2016). However, these expressions are inconsistent with substantial uncertainties with various perturbations such as earth gravity, atmospheric drag, multi-body gravitation, solar radiation pressure, or longer propagation time (Park, 2016; Shou, 2014).

Later, researchers proposed an analytic solution to solve the problem. Many researchers have explored analytical and semi-analytical solutions to explain the

orbital motion with additional perturbation values (Park, 2016). In addition, analytical and semi-analytical solutions can provide an in-depth view of classifying variations in an object's orbital motion. This fact inspires a new perspective to explain uncertainty's evolution and implies new orbit propagation methods.

Machine learning approaches present different modelling capabilities and forecasting than physics-based methods (Gonzalo and Colombo, 2021). The forecasting process can be performed without explicitly modelling space objects and limited space environment information. Instead, the models are studied based on the observed data. There are various types of machine learning approaches. The most common types are; supervised, unsupervised, and reinforcement learning (Jo, 2021; Abu-Mostafa *et al.*, 2012). Supervised learning is a method that studies the function or mapping of labelled data (Xie and Huang, 2021; Huang *et al.*, 2014). Yet, unsupervised learning is a learning method from unlabelled data (Wang *et al.*, 2021; Novotny *et al.*, 2018). It uses to find patterns and structure data such as cluster data into different groups without providing output to describe groups. Besides, reinforcement learning is used to make decisions (Leng *et al.*, 2021; Abu-Mostafa *et al.*, 2012). Thus, supervised learning is ideal for increasing orbit propagation accuracy based on historical measurements. In addition, it is more straightforward and not complicated compared to other methods.

Also, using the data-driven approach for orbit propagation allows the prediction process through data processing to aid decision-making (Jiménez-Luna *et al.*, 2021). The data-driven approach can produce a precise orbit propagation model, although there are elements that mathematical models cannot determine. For example, identifying perturbation value through data approaches (Jiménez-Luna *et al.*, 2021; Jäggi and Arnold, 2017). Peng and Bai (2017) used the data-driven approach to reduce orbit prediction errors from historical data and increase orbit propagation accuracy.

Recently, the hybrid propagation method has improved the accuracy of the orbit propagator (Lopez *et al.*, 2021; San-Juan *et al.*, 2017). This method combines classical integration methods with a forecasting technique based on either statistical time series or machine learning techniques. It can model the difference between the

integrated solution and natural behaviour. The hybrid propagation method also enhances orbit propagation by refining the analytic approach and improving computational efficiency (Park, 2016). San-Juan *et al.* (2017) improved 90.03% over 30 days of the SGP4 Model by using this hybrid method with statistical technique. However, it also causes complexity and computational burden for the end-user (San-Juan *et al.*, 2017). Later, Peng and Bai (2017) improve orbit propagator accuracy by up to 96.1% using a support vector machine (SVM) for medium-term forecasting. Therefore, further study is required to explore the recent method used and appropriate learning techniques to accomplish this study.

### **1.3 Problem Statement**

The SGP4 Model is the orbit propagator model commonly used and known as the most advanced space surveillance system. However, this SGP4 Model has an error of  $\sim 1$  km and grows at  $\sim 1-3$  km per day. Besides that, the TLE data used in conjunction with the SGP4 Model for space operations can be forecasted with accuracy valid for 2- 3 days. It needs to be updated periodically to minimize the error. The increased propagation span caused the position error to rise to 270.7km for 30 days. Thus, the operation planning will disrupt and create problems for the user, such as the desired image cannot be taken due to an incorrect satellite position. In case of collision, accident detection may be too late for prevention. Recently, the hybrid propagator method has improved the orbit propagator. The hybrid propagator method extends the validity of TLE data and reduces the error in the SGP4 Model. The Holt-Winters technique was used in this hybrid propagator, and it improves 90.03% over 30 days of propagation. However, this method requires changes in the components' probability distribution, causing complexity and computational burden for the end-user. Later, the SVM can reduce error and improve orbit propagator accuracy up to 96.1%. However, this technique's correction capability is limited and adequate after 28 days. Its ability reduces for long-term forecasting. In the literature, it has been found that ANN has the most regression capability due to its more flexible structure. Thus, this study explores deep learning techniques as they are scalable and suitable for complex data. The Recurrent Neural Network (RNN) and long short-term memory (LSTM) are deep

learning methods ideal for time series forecasting and can remember patterns for long-term forecasting. Therefore, an improved orbit propagation model can be developed through the hybrid of the RNN-LSTM with the SGP4 Model. This hybrid RNN-LSTM is proposed because integrating these two techniques can assist in long-term forecasting with minimal error as they complement each other and overcome their weaknesses. Also, it can reduce the reliance on external inputs in performing forecasting. It also provides a solution to sequence and time series-related problems. Furthermore, the forecasting only required initial TLE data for long-term forecasting. Thus, it avoids relying on updated TLE data and enhances the current SGP4 Model capabilities.

#### **1.4 Research Aim**

This research project aims to improve the SGP4 Model for orbit propagation using the deep learning approaches. As a result, the improved SGP4 Model will overcome the limitations of orbit propagation for long-term forecasting and handle complex satellite data information. Furthermore, the enhanced SGP4 Model will validate. Hence, the results can be evaluated and discussed.

#### **1.5 Research Questions**

A set of research questions is formulated to find the solution for this research study. The following are the research questions used to guide the research accomplishment:

- (a) **(RQ1):** What are the essential features in the space object data to form the SGP4 Model?
- (b) **(RQ2):** How to design the SGP4 Model framework for identifying elements involved in improving the SGP4 Model?

- (c) **(RQ3):** How to develop a hybrid RNN-LSTM SGP4 Model for minimizing errors and maintaining accuracy in long-term forecasting?
- (d) **(RQ4):** How to validate the improved SGP4 Model based on error rate forecast measurements?

## 1.6 Research Objectives

Following are the research objectives of the study to achieve the aim of the research:

- (a) **(RO1):** To identify essential features in the space object data for pre-processing to form the SGP4 Model.
- (b) **(RO2):** To design the SGP4 Model Framework in identifying the elements involved in improving the SGP4 Model.
- (c) **(RO3):** To develop a hybrid RNN-LSTM SGP4 Model for minimizing errors and maintaining accuracy in long-term forecasting.
- (d) **(RO4):** To validate the improved SGP4 Model based on error rate forecast measurements.

## 1.7 The Scope of The Study

The research is bounded by:

- (a) The space object data used in this study is a Low Earth Orbit (LEO) space object data. The selection of The LEO space object for this study is because it is this space object that is tracked by the Ground Station, Malaysia Space

Centre. Therefore, the results of this study will be able to assist the monitoring and operation of the satellite operators. In addition, LEO space object data is the most abundant data in orbit. Its increasing number poses a danger to the surroundings because they can collide, and its position is close to the Earth compared to other space objects. Therefore, LEO space object tracking is crucial compared to other space objects, such as in Medium Earth Orbit (MEO) and Geostationary orbit (GEO).

- (b) The forecasting analysis is focused on the learning-based process approach. This approach is used because it is less complicated than other approaches, such as the statistical-based approach or mathematical approach, which requires mastery of mathematical formulations that are more complex and complicated. In this study, a learning-based approach that uses machine learning techniques through deep learning can optimize the use of historical data. The method is also less complicated and suitable for dealing with the development of a long-term forecasting model.
- (c) The space object data used in this study is text type. The selection of text type data was made because the forecasting results for this study will also be in the form of text. In addition, the enhanced model developed is a regression model, and text-type data can assist in long-term forecasting. Although there are other types of space object data, such as images, for this study, such data will not be used and will be unsuitable for solving the long-term forecasting issues.
- (d) The algorithm used in this study will also focus on appropriate techniques for time series data only. This is because the data features selected to be trained in this study, namely position and velocity space object data, are time-series data.
- (e) This research does not investigate the privacy issues or the security issues related to orbit propagation. These privacy and security issues include tracking a spy satellite used by certain parties for intelligence purposes, warfare, and so on. Therefore, the results of this study will only help the parties involved in terms of governance of space object data only.

## **1.8 The Significance of the Study**

The significance of this study can be gained through practicality and theoretically as follow.

### **1.8.1 Practicality**

#### **(a) Nation**

The study will play an essential role in the space industry and assist the regulatory authorities in monitoring the space asset and public safety. In addition, it will provide additional inputs to the international regulatory authorities monitoring space object issues such as NORAD and other related international organizations. Thus, it will prevent unwanted events from happening. In turn, it helps speed up the prevention process in any incident.

#### **(b) Space Agency**

The study's outcome will allow the space agency that cannot afford the related facilities to benefit at a worthwhile cost. Subsequently, it will assist in administering the operational management of the ground station activity. As a result, space-based organizations can have their orbit propagation system without relying on other space agencies. Also, this orbit propagation model system can be designed based on their requirements and needs.

#### **(c) Researcher**

The ideas presented may be used as reference data to conduct new space data analytics research and improve current research. The researcher or analyst can make the right decision based on the deep learning approach despite their lack of expertise and facilities.

- (d) Individual

The study will produce a reliable orbit propagation model and provide vital information for individuals or anyone affected without gathering the information from the authorities or any related organization.

### **1.8.2 Theoreticity**

- (a) The study helps to improve the method for the long-term forecasting horizon of space object data.
- (b) The improved model is more robust and resilient to the effects of the space environment as the modelling does not require the latest space environment data as it is based on the historical data used.
- (c) The improved model framework can assist in having systematic rules when processing and preparing the raw space object until the data is ready to be used for the modelling process.
- (d) The improved model framework can assist in future modification of the orbit propagation model and dealing with space object data.

### **1.9 Thesis Organization**

The thesis's organization is structured as follows;

- **Chapter 1** presents the research work's introduction, describing the research's overview and motivation, the background of the problem, problem statement, research aim, scope, and significance of the study. The problem statement highlights the need for an orbit propagation



model for long-term forecasting. Also, to provide an accurate result-based deep learning approach to overcome the current SGP4 Model limitation.

- **Chapter 2** reviews related works and evaluate the previous study and results that led to the research gap. Then, based on the literature investigation, the way forward of the research is discussed.
- **Chapter 3** describes the research methodology and deliverables for the activities involved in achieving the research aims. The research plan contains five (5) phases that are: 1) Preliminary Study, which aims to analyse the problem and limitation of the current orbit propagation model and identify an approach that can improve the SGP4 Model's performance; 2) Data Collection, which explains how the collection of data is executed; 3) Data Analysis and Pre-Processing, which to pre-process the space object data; 4) Design and Modelling which to design and model-based deep learning approach to overcome the SGP4 Model limitation which is the integration between the SGP4 Model and the RNN-LSTM module, and 5) Validation Phase which aims to validate the improved model through experimental simulation and analyse the performance in terms of error rate forecast measurement.
- **Chapter 4** presents the data analysis and pre-processing for the study. Firstly, the analysis of data is conducted. Secondly, the space object data collection is explained. Next, the space object data is studied to understand more about the data. Then, the execution of pre-processing of the data. A suitable sampling interval is proposed based on the analysis conducted. Then, identify the essential features in the space object to form the SGP4 Model, and present the behaviour of the data patterns. This chapter addresses the first research objective (**RO1**) of the thesis, which identifies the essential features in the space object data for pre-processing to form the SGP4 Model. With that, this chapter provides the first contributions of this study.

- **Chapter 5** explains the design of the SGP4 Model Framework. This chapter also addresses the thesis's second research objective (**RO2**), which is to design the SGP4 Model framework to identify the elements involved in improving the SGP4 Model. In addition, this chapter contributes to the second contribution of the study.
- **Chapter 6** addresses the third and fourth research objectives (**RO3**, **RO4**) to contribute to the thesis. First, it will explain the RNN-LSTM module development contribute to the integration model's development. Next, the integration model is validated to ensure this study's accomplishment. Thus, this chapter contributes to the third and fourth contributions of the thesis.
- **Chapter 7** addresses the last research objective (**RO4**) of the thesis. On top of that, it will elaborate the experiments' results and discussion of findings in detail. Thus, this chapter contributes to the final contribution of the study.
- **Chapter 8** summarizes the achievement of the thesis, the research contributions, limitations of proposed approaches, research outcomes, and future works.

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## LIST OF PUBLICATIONS

### Indexed Journals

1. **Salleh, N.**, Yuhaniz, S.S., Sabri, S.F., & Azmi, N.F. (2020). Enhancing Prediction Method of Ionosphere for Space Weather Monitoring Using Machine Learning Approaches: A Review. *International Journal on Advanced Science, Engineering and Information Technology*, 10, 9-15. **(Indexed by SCOPUS)**
2. **Salleh, N.**, Yuhaniz, S.S., Sabri, S.F., & Azmi, N.F. (2022). Modeling Orbital Propagation Using Regression Technique and Artificial Neural Network. *International Journal on Advanced Science, Engineering and Information Technology*. **(Indexed by SCOPUS)**

### Indexed Conferences Proceedings

1. **Salleh, N. A.**, Yuhaniz, S. S., Azmi, N. F. M., & Sabri, S. F. (2019, February). Enhancing Simplified General Perturbations-4 Model for Orbit Propagation Using Deep Learning: A Review. In *Proceedings of the 2019 8th International Conference on Software and Computer Applications* (pp. 27-32). **(Indexed by SCOPUS)**
2. **N. Salleh**, N. F. Mohd Azmi and S. S. Yuhaniz, "An Adaptation of Deep Learning Technique In Orbit Propagation Model Using Long Short-Term Memory," 2021 International Conference on Electrical, Communication, and Computer Engineering (ICECCE), 2021, pp. 1-6, DOI: 10.1109/ICECCE52056.2021.9514264. **(Indexed by SCOPUS)**